

## 

- Aerospace Engineering -

#### **Numerical Method**

- 3D Simulations are carried out by the solving Navier-Stokes equations using a finite volume scheme that is second-order accurate in space and time which employs Large Eddy Simulation (LES) approach.
- Reaction rates are modeled by using the Linear Eddy Mixing (LEM) and the Eddy Break Up (EBU) model.
- The chemical kinetic mechanism used comprised a single step, global mechanism for heptane/air combustion which involved 4 species ( $C_7H_{16}$ ,  $O_2$ ,  $CO_2$  and  $H_2O$ ).
- The liquid jet in cross flow is modeled by means of a Lagrangian approach. Breakup approach is not yet considered in this simulation.

# Simulation conditions

- Inflow
  - Mass flow rate : 0.1596 kg/s
  - Static temperature : 873 K

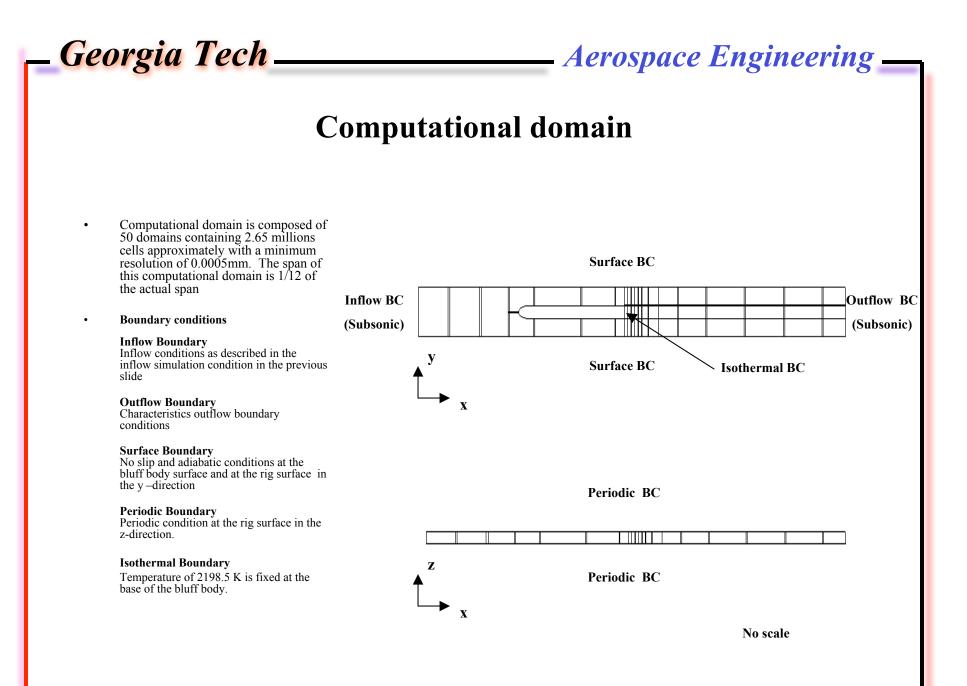
<u>- Georgia, Tech</u>\_\_\_\_\_

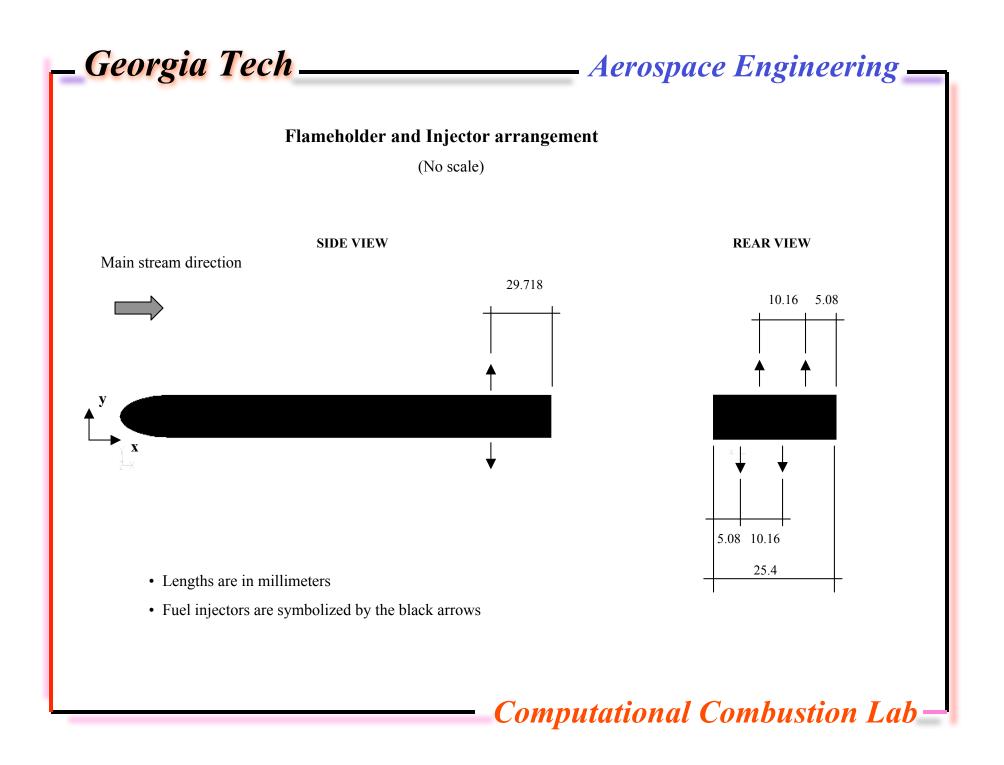
- Static pressure : 111350.33 Pa
- Velocity : 126.68 m/s
- Inflow composition
  - Y<sub>C7H16</sub> : 0.000000000
  - Y<sub>02</sub> : 0.1745032727
  - Y<sub>CO2</sub> : 0.0480192657
  - Y<sub>H2O</sub> : 0.0224646204
  - Y<sub>N2</sub> : 0.7550128412

- **Fuel Injection** 
  - Heptane mass flow rate :  $m_{C7H16} (\Phi_{overall} = 1) = 8.536267 \times 10^{-3} \text{ kg/s}$

\_\_\_\_\_ Aerospace Engineering \_\_

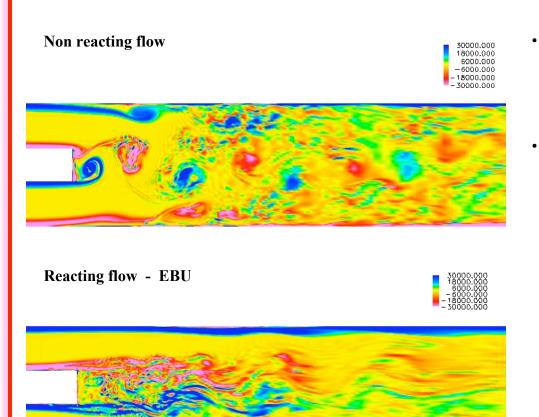
- 4 injectors of radius 0.50 mm
- Velocity of injection : 15 m/s
- Injection Temperature 286.11 K
- SMD radius 30x10<sup>-3</sup> mm
- Sigma = 0.5
- Cooling
  - Air flow rate : 0.110 kg/s
  - Temperature 348.89 K





## - Aerospace Engineering \_\_

#### Results



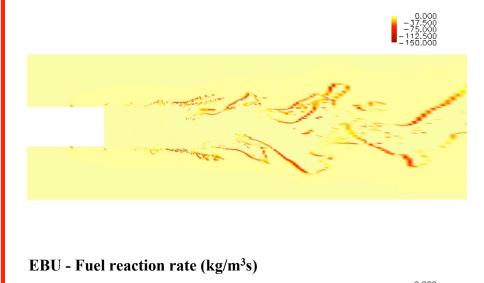
<u>\_ Georgia Tech \_\_\_\_\_\_</u>

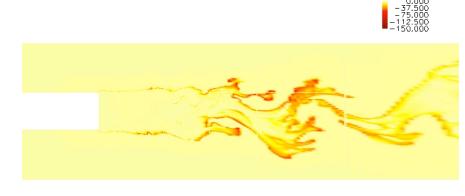
- The flow solution at the at a z = const plane corresponding to the middle plane of the rig.
- Results are reported for a simulation duration of 1.4 flow through times – not strictly reached statistical stationarity but both LEM and EBU studies have evolved a similar extent to carry out qualitative comparison
  - The non reacting case is dominated by asymmetrical vortex shedding as evident from the Von Karman vortex street.

Aerospace Engineering \_

#### LEM - Fuel reaction rate (kg/m<sup>3</sup>s)

Georgia Tech \_\_\_\_





- The recirculation zone at the base of the bluff body is a source of heat which allows the combustion process to occurs in which . the flame is stabilized. However, the flame exhibits an unsteady pattern
- The LEM predicted flame structure is much thinner and located in the high shear regions whereas the EBU flame shows more local thickening. The LEM prediction is a subgrid model using finite-rate kinetics coupled with molecularl diffusion and turbulent mixing and the profiles are the LEM-filtered fields.
- Downstream of the nearly symmetric flame region, the flame exhibits a sinuous pattern due to the vortex shedding.
- At the region where the vortex is shed, local extinction may be occuring. At these locations, the flame is broken creating regions of hot products which are convected and react with unburned mixture downstream.

